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MAY 17, 1926

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NUMBER  
20

METAL CONSTRUCTION OF AIRCRAFT  
THE PITCAIRN FLEETWING  
MILITARY AVIATION BILL PASSES HOUSE

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# AVIATION

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## AVIATION

VOL. XX

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No. 20

### A Great Achievement

LEFT COMMANDER BYRD'S flight to the North Pole is one of the great epics in the development of aviation. Nature put up barriers of ice and snow around the Pole which made it almost impossible to the scientists of man. Careless expeditions with ill-fated and ill-equipped have struggled toward the Pole only to be tossed back and forth by the terrible cold and the roaring, snow-filled wind. Misadventured and the north of science seemed to be all at sea in venturing from the shores of the North. Every likely vessel the Pole through sheer determination of purpose and, through the use of better strength applied to the most primitive equipment. Nothing can do this achievement, but the conquest of the polar regions did not increase their accessibility to man. It took Peary eight months of almost superhuman effort to reach the Pole and return. It took Byrd less than a day. The conquest is most complete and extraordinary. The barriers to Northern exploration have at last been broken down. The voyage has again shown that it can do things which no other form of transportation is able to accomplish. It should be pointed out that the flight was either easy or that at the present time it can be repeated without an element of considerable risk, but it has shown that all parts of the North can be reached in a minimum of time by the route of the air. Ultimately it may lead to the development of air routes between the Americas and Asia, certainly by way of the polar region, but this is still far in the future.

The flight which Byrd accomplished this year would not have been possible a few years ago. The difficulties of navigation at last year and the American flight both contributed to the possibilities of success. The development of observation instruments which can be used from the air is a very recent science and one to which Byrd himself has made great contribution. From the astronomical viewpoint the flight is a complete validation of the value of the three-legged plane and of the ability of the air-model engine in function under most adverse conditions.

Byrd's flight to the Pole, like the flight around the world, will keep the interest of the public focused on aviation and will increase his belief in the possibilities of aerial transportation.

The newspapers are giving a large amount of space to the flight, and as one reader will already have had all the news before the news reaches them, we will not try to duplicate the general story. In a later issue, however, Commander Byrd's flying experience and technical observations will be covered.

Amateur aviation extends its most sincere congratulations to Commander Byrd and Pilot Floyd Bennett for having successfully accomplished what so many have dreamed of doing.

### Cooperative Flying Clubs

THE IDEA back of the English "Light Plane Club" could well be followed in this country. There are, undoubtedly, a large number of young men who want to fly but are unable to do so on the expense of owning their own plane is too great. If they could have the use of a plane at reasonable rates they would do a considerable amount of flying. The difficulties of joint ownership and use are considerable, but they have been solved in certain spheres, such as the athletic and recreational facilities offered by many clubs, and they could undoubtedly be solved in regard to the joint use of a plane.

Actually, the idea of a co-operatively owned plane could now be put into effect in many towns. Many of the members of Aero clubs or of numerous chapters of the N.A.A.U. are sufficiently wealthy and interested in aviation to subscribe the capital price of the plane, if these individuals were set in a position to make use of the plane themselves. If such a plane were developed, the club could afford to build its own hangar and keep a pilot and mechanic and in most cases it would probably be better to leave the maintenance of the plane to the local flying company. This would save much work on the part of the members of the club and would also help out the particular aerial service company. The arrangements and the rates charged would have to depend entirely on local conditions and could be worked out in many different ways, but, undoubtedly, flying could be greatly encouraged, especially if the non-flying members paid part of the expense of the club.

Flying for sport depends a great deal on the atmosphere in which it is done. A group of congenial friends who club together and own a plane of their own can have a great deal more fun, and therefore, will fly considerably more than if they had to go out and rent a plane. The method by which such a club would be started would differ with different localities. In some cases it might be fostered by a man, it would be started by a group of flying men, in other cases, it might be organized by the local air club. In almost all cases there would probably be only one or two people who would be willing to put the thing over. The professional pilot of the locality could spend his time profitably by working out the cost of the flying, arranging for the maintenance and storage of the plane, etc. As the primary object of the club would be to give reasonable rates to its members the pilot should not expect an undue profit. If the club were well handled, he could expect a considerable increase in the amount of local interest taken in flying. He would be giving instruction and, in the long run, would all potentia places to the members of the club and often make valuable financial contacts if he wished to expand his business.

# Metal Construction of Airplanes

## 4 Detailed Discussion of the Conversion of the Navy TS Plane into the Dunscombe Built F4C-1 Plane. The Principles of and Reasons for the Cambered Spar.

By CHARLES WARD HALL

The metal construction of airplanes is a subject of an extremely specialized nature, the problems of which are of very first importance since the type of construction is becoming more and more general and is favored by both service and commercial aircraft construction. In the field of development, few have given closer attention to the problem involved than has Charles Ward Hall, of Charles Ward Hall, Inc., 136 East 42nd Street, New York City, N. Y. In addition to a number of original airplane designs, carried out in all-metal construction (with fabric covering, including the Hall-situated sport plane, the company has developed a number of conversions to the sport order of the Dunscombe Variable in the class, in the Navy F4C-1, a single motor fighter, which has been adapted, in the main, by Charles Ward Hall, from the Navy TS plane. The work was carried out late in 1934. The Curtiss Dunscombe and Motor Co. Inc., designed and forwarded the following, in connection with the production of the plane: The wheel type undercarriage, engine cooling, installation of fuel system, and machine guns, etc. It should be pointed out that, while the designation of the F4C-1 indicates it to be a single-engine fighter, the production, the TS plane, was classified prior to its adoption of the Navy's new standard system of plane designation and, consequently, bears an obsolete designation. The metal portion of plane designation in art (other alterations in the case of *Aviation*—Editor).

The F4C-1 airplane is a biplane of 25 ft. span, 19.23 ft. overall length and 8.75 ft. extreme length, with a gross weight of 1867 lb. light and 2700 lb. loaded. The machine is a metal version of the standard Navy TS plane which was produced in wood. It is designed to be replaceable either in a land type undercarriage or to the attachment of two floats as a seaplane. The power plant, tail assembly, structure and control load frame are identical in the TS and the F4C-1.



A ground view of the F4C-1 Navy Fighter (Wright Whetstone). The apparent sagging of the rear part of the upper wing is due to the combined forces of the span which is fully stressed in this article.

except that the F4C-1 has a metal propeller weighing 27 lb. more than that in the TS machine. Likewise the power plant and construction, as a unit, weigh 37 lb. more in the F4C-1 plane than in the TS design. The wing section is identical in each case. The bend dimensional details of the TS design are as follows: Span, 25 ft.; overall length, 21.4 ft.; height, 9 ft.; weight empty, 1226 lb., and weight loaded, 1965 lb. Because of the close similarity between the two designs and the equal metal loads, it will be interesting to discuss, at some length, the structural details of the metal plane.

### All Metal But Lighter

The F4C-1, except for the fabric covering of the wing and tail surface and of the fuselage, is constructed wholly of duralumin, whereas the TS, also fabric covered, is built entirely of wood, with the steel bracing, in general, metal framed engine mount as much as, and frequently more than, their counterpart in wood. In the case of the F4C-1 and TS designs, however, the gross weight of the complete metal airplane is much less than the wooden machine. In fact, the weight of the bare structural frame of the F4C-1 is less than one-half that of the TS.

Then very considerable saving of weight, therefore, cannot be attributed solely to the more substitution of materials. It is largely due to a thorough system of design, not merely involving the correct proportioning of the principal members, but recognizing the stress patterns to which they may individually be subjected, but extending to even comparatively minute details.

As a preliminary, a very complete stress analysis of the entire structural framework was made, including an investigation of the effects of redundancy and of the distortions due to the loadings, the familiar method of least work not also the true commonly applied law of suspended deflections (usually attributed to Maxwell) were used wherever appropriate.

principle, secondary stresses were treated exactly, (a), in such a manner as to balance one against another wherever practicable, thus reducing the effect of both; (b), by introducing secondary stress to relieve the first stage of apex hollow sections, or to control the places of unsymmetrical distortion; (c), by maintaining very gradual changes in length, width, or moment of inertia, in the members of any truss; (d), by proportioning the various trusses to enable single construction and to form approximately equal

considerable bending stress combined with compression, as in many cases, the single tube is constantly heavier, and just as often that is a truss form of construction. In such a spar truss, if the chords are formed of a plurality of tubes and the web members are also of tubular form, such a proper series of apex depth to apex width in relation to the length of the W legs and the deep legs, is maintained, the almost possible efficiency attainable by more form of cross section will result.



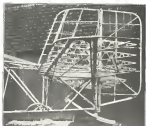
The upper wing panel seen from the underside lateral viewing.

lateral triangles, so far as practicable, short panels were arranged at random must choose in compression, (e), in the use of fuselage slaying bars and stiffeners while also a large ratio of length to width, (f), in arrangement of each web member so as to form an integral part of the entire truss, and (g), by dissuading secondary stresses where such could not be obtained by subtractions, in some cases, by the use of fully composite joints, while, in other cases, by maintaining the curvature of the gravity axis of members subject to combined side load and end load so as to eliminate the effects of misalignment, etc. The effective strength of members has been enhanced through treatment by the fitting of their ends. This has been done in nearly all cases where the distortion of the stress lines under load would thus result in double curvature of the member. In cases where single curvature would result, and bending has been avoided.

### Wing Ribs Formed of Single Sheet

Surprisingly of construction has rendered considerable efficiency. As an example of this, the wing ribs and tail surface ribs are integrally formed from a single sheet. Likewise, fuselage joints, in general, consist of a single forging, including, fitting the hangers and provided with integral legs rigidly secured to the web struts and stays. Whenever practicable, round tubing has been used for structural members. In those cases where built-up tubes the form of section, the most efficient available form has been used. For resisting torsion or compression, a single tube of symmetrical thickness and diameter is, of importance, 1895 efficient. Where there is

distortion, some 40% or more, may be added in the ultimate stresses of stress on a unity per foot of spar, by shearing it in a vertical line between the supports, instead of bending it straight, and by reducing the cross-sectional area of the chords at their less stressed parts. Such a member is a structural truss, the stress of the curve coinciding with the deflection stress, in supports, with the length being nearly uniform where supports. Redundancy of the area of the chords, where there are less stressed under the combined side and end load, involves, for an equal spar, the reducing of the lower chord area in the supports and of the upper chord area

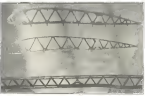


A ground view of the tail section of the F4C-1.

near and open. For a lower spar, under such combined load, the upper chord is reduced at the supports and the lower chord near the open. Similarly, such a variation in area between low upper and lower chords results in a shifting of the stress axis of the spar section in a position every time the position of symmetry and axis to the lower chord. In other words, the actual gravity axis of the spar has been considered about the open length of the spar because so of the actual structure of the spar itself had been carried.

### The Cambered Spar as Practice

When an upper spar in an airplane wing has been so constructed and the plane is in the condition of pulling out a turn, the lift load tends to straighten the spar between the airfoils and the wing length, if the spar has been correctly proportioned it will be quite straight when the finished result.



In the upper part of the photograph a ground view of the wing is shown in the wing spar, while the same of, after passing through the turning due, is shown below. The spar is a curved section of an open rib. In the lower part of the photograph, a section of a spar which has undergone an extreme but of combined end and side load.

mean load factor has been reached and, then, the ultimate loading moment due to the product of end load and the compressive force in a normally compressed wing which would be dissipated under the load, will be dissipated.

In reinforced joints of this type, the ultimate load sustained, as shown by tests, is not at the yield point, as would seem to be indicated by the permanent set after the removal of a load increment. On the contrary, these joints will sustain, before failure, 40% to 50% more load than was required to produce the first yield marking set.

#### Preventing Column Action Failure

Another method of preventing failure by column action through the shiffling of the joints also, consists in evidently varying the width of the members, as is done with the ribs members of the rib truss. These are of parallel cross section and approximately 55 deg angles. The apex is near fixed straight and the ribs are fixed at its apex to the wing skin, subjected to order to transfer load on the fixed flange. This arrangement results in a gradually reduced static axis of the ribs away from the apex, as load is applied, to increase the compressive force at the apex of the section and to reduce the stress as they distill, do not buckle.

Another type of reinforcement against buckling is used for the ribs where the outer flanges of which are reinforced



A typical method for applying the filar sheet to outer ribs.

around a joint point. Such bracing adds about 60% to the ultimate strength of the entire rib. The ribs, as a whole, are of X section and asymmetrical. Instead of setting them in a plane perpendicular to the wing panel plane, they are set at an angle of 90 deg there, thereby absorbing failure.

#### Eliminating Secondary Stress

Wherever, as is found in tapered columns, the tapered web system introduces in a rectangular panel plate, the secondary stress in the chords adjoining the joint is very large, and subsequently 100% of the primary stress. This is due to the widening of the column under load, which accompanies its shortening. The tapered web system has no tendency to restrain this widening but a rectangular panel effectively provides it and so produces a large bending moment in the half panel adjoining. The panel joint plates of these trusses, of Y form, eliminate any half panels, and permitting the attachment of left and right bracing to the spars without interruption of the continuity of the diagonal rib truss. The elimination of the large secondary stress, otherwise present, allows the use of lighter members for the spar chords. Incidentally, the weight of the joint plates is also greatly less.

The drag bracing throughout the wings of the F4C consists of tubular struts provided with externally capped end fittings, which engage the upper surface of a double half head bolt. The drag wires are attached to a forked and externally capped strut, the latter being connected to the lower half

surface of the bolt which is passed through drilled holes in the drag and spar panel plates, and secured by a nut. Important elements of strength and of weight saving result from this simple design. The deep capping of the strap and the widely separated points on the rim of the nut from which the strap flange bend, cause the line of action of the drag wires to intersect the axis of the bolts at their bearing



On the right is shown one end of a drag wire on the wing structure with the filar for the drag wire. On the left is a typical drag wire joint.

on the right. Tests have shown this arrangement to fail at ultimate load by the shiffling of the bolt around the perimeter of the bearing, indicating the shiffling of a bearing moment rather than the joint to open at the drag wire connection. As a result, the spar joint plates are less free between and they, as well as the spar chords, may be tapered.

#### Creating a Friction Drag Moment

The bearing at the drag strut fitting upon the upper surface of the half struts on load is applied, in preventing these ribs from buckling, not from the resulting friction, a drag moment exists. This bearing moment is caused by the pressure of the end load, and is, in fact, necessarily equal to the product of the coefficient of static friction, the radius of curvature, and the end load. The drag moment for any end load one thus be computed definitely. On tests of the drag struts and on the F4C-1 wings, this frictional force moment was reduced by order of 8 factor to the static formula of  $3 \times 10^3$ , the tubing being only commercially straight and not specially rolled. This is a higher factor than can readily be obtained by testing with carefully made 4 in. radius bearings, each long struts (10 ft minimum) being perfectly straight, up to the fitting load, and when the section is excessive, drag wires suddenly to a curved form, and increase to orders a factor factor of 1 while no less



A tapered rib section after being tapered a end load of 150 lb per ft. The simplicity of design of the outer structure is characteristic.

When the load is removed the struts become straight and may be reloaded repeatedly with the same result. The struts so loaded or "zero loaded" are now in a light. Another somewhat method of creating end filar is a number of struts running of one point on the fitting and for the fittings. The struts and struts which form the web system of a Warren truss are constructed of tubing, a line of tubing about three diameters long being first secured into each end. The tube ends and bases are fabricated in a special

designing tool which cuts first by a press fit upon the first leg of the fitting. The struts are fastened to the legs by means of two staggered rivets, thus providing a structure which, under test, developed around every fiber in the Kelly formula of about 2 for the bearings and 1.5 to 2.0 for the struts. The end ends of such a bearing is usually determined by the thickness of the fit in the cylindrical part of the fitting and, normally, by the length and stiffness of the base struts or struts within at one point. As at least one, and the

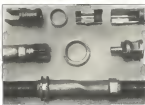


Fig. 1. End fittings for drag and bracing tubes, characterizing the design of wing ribs.

spare parts of the struts and the outer bearing, an additional stiffness is obtained in the combination. The end ends of such struts depends chiefly upon the thickness of the fitting leg, and is, therefore, very readily comparable to usual cases.

#### Saving Weight in Connections

Further strength of weight saving can be obtained through particular attention being given to the methods adopted for connecting the principal members together while, at the same time, relieving their full strength as shown in the photograph, Fig. 2. The upper outer portion of this photograph shows a section through the joint of two tubes, each by meeting into the center tube a row with an angular disposition and

into that above in the upper left corner of the photograph, and then by telescoping it into a larger tube, outside of which is a narrow ring, such as is shown. The outer ring is then compressed, with the two tubes, into the depression of the inner fitting. In the photograph, the finished fitting is shown in the middle row on the left, while a section of this fitting will be seen in the top row on the right, the fitting and ring being completely assembled showing about the proportion of these above in the upper row on the left. Such connections, properly proportioned, develop the full strength of the tube in compression, in tension or in bending, and, when failure occurs, it is definitely remote from the joint. By joining the inner fitting to a polygonal cross-section as shown in middle row, right, of the photograph, and by compressing a tube and an external ring around it in the same polygonal form, a joint is produced which is no slower at the bottom of the photograph, 300% increase in tension.

#### Feasibility of Ring Joints

Joints of this type are not limited in production to very ductile material, in weight or speed, but are readily produced in material having a specified tensile strength of 50% or less in a variety, though approximate, relation between the thickness of the tube, the curvature of the filar and the depth of the member depression, and between the diameter of the tube and the length of the ring. The tube, inner fitting and ring, of the various shapes, were laminated and tested, open and after before the sections were photographed in order that the parts might show clearly the details of construction. Actually, such a section shows in joint tests engaged in 20 or more diameters.

This method of construction has been very thoroughly tested. The whole system of design is the result of extensive research into the most efficient and the lightest method of supporting loads, compression with adequate strength. Several thousand qualitative and quantitative tests were carried out in the development of the various methods of construction employed. As a result, the F4C-1, as an example of the practical application of the various methods, has proved its adequacy and complete strength under all conditions.

The F4C-1, and other airplanes produced by Douglas, Ward Hill, Inc., is a structural variety of developments, with the exception of some of the labor covering and any bearing wires. It demonstrates a method in which, based on skilled without disturbing its essential properties, the function of



The complete structure (shown in assembly) of the F4C-1



# The Madrid to Manila Flight

Second Long-Distance Spanish Flight Not Successful on Previous South Atlantic Undertaking.

PERMANENTLY DISCOURAGED by the almost catastrophic success of the recent Spanish flight of Madrid-Buenos Aires from Spain to the Argentine, another long-distance flight was immediately ordered and on April 8, three Hispano XLIIIs, equipped under the Spanish flag, set out from Madrid for Mexico, Palmyra Island, the Philippines and Manila. The pilots of the planes were Capt. Hector Martinez-Rodriguez, commander, Capt. Joaquin Lopez-Delmon, and Capt. Eduardo Gomez-Godoy, while the mechanics were Pedro Mariano Garcia, Joaquin Arcones and Eugenio Perez. The order of the flight was laid out as follows: Madrid, Algeria, Tripoli, Cairo, Egypt, Damascus, Syria, Baghdad, Iraq, Persia, India, Bombay, Penang, Bunder Abbas, Penang, Rangoon, India, Calcutta, India, Bangkok, Siam, Saigon, Hong Kong, Hong Kong, Manila, Luzon, and Manila. The planes, in addition, each of French design, were constructed in Spain, and the squadron was given the name of 81. Once after Juan Sebastian del Cano, who brought news of Magellan's voyage back to Europe and thus may be claimed as the first circumnavigator of the globe. The mechanics are equipped with single Latécoere-Dornier engines of 450 hp and each plane carries 355 gal. of gasoline and 25 gal. of lubricating oil.

## The Start

The squadron left on its flight at approximately 10,000 miles at 8:45 A. M. on April 8, over Central Vietnam territory near Madrid. The first stop was planned to be in Algeria, Tripoli, North Africa, a distance of about 800 miles. The planes passed over Cartagena at 10:30 A. M. and then headed toward the sea in the direction of the African coast.

They reached Algeria successfully and without apparent incident. On the next day (April 9) the three airplanes set out for Tripoli and Captain Rodriguez, commander of the flight, landed at Tunis, North Africa, leaving the right of the other two planes in a fog. The other two mechanics provided strength to Tripoli and landed there in a few minutes. Captain Rodriguez, alone continued a head-on run on the right wheel as heading bad, having reported that the flight was recovered the next day to Tripoli and on each to reach the other two planes. The flight was, undoubtedly, in fact, weathering the rain and was being very dark. In accordance with a previously arranged plan, the other two planes did not land but continued on to Tripoli and proceeded to Bengasi, Cyrenaica, on April 10, returning to the next day in Cairo, where they landed at the Royal Air Force airfield at Helwan. In the next hour, Captain Rodriguez had reached Bengasi, between

them, only a few behind his companions. However, at Cairo, Captain Rodriguez and Godoy landed in arrival on April 10, and work was commenced on overhauling the three planes.

Recent trouble did not overtake the three until the flight from Cairo to Baghdad, Iraq, was undertaken, on April 11. The three mechanics left Helwan at approximately 7:00 A. M. On the flight, the planes got separated and Captains Lopez and Godoy landed Baghdad at 4 P. M. and 7 P. M. respectively, while Captain Rodriguez' machine did not arrive and has not been heard of. The distance between Cairo and Baghdad is about 800 miles, across the Syrian Desert. One of the other planes was forced to land in the desert to repair a fuel tank, according to dispatches, but around the flight.

The flight immediately set out for H. A. P. planes to search the desert for Rodriguez who was last seen very early from Amman after having passed Damascus. The route between Cairo and Baghdad which is frequently forced over by the British service pilots, is well marked, there being a large map laid straight across the surface of the desert for the guidance of pilots. It is said that Rodriguez passed within "10" on the route and then drifted with his companions in the desert, unable going off as a perfectly accurate. His companions, believing his machines, did not follow and as a result of the leader was discovered until the mechanics found his machine on April 13, 140 miles from Amman. The plane appeared to be in perfect condition and a note was attached to it, explaining that the pilot and mechanic had decided to make their way on foot in Amman. The landing was apparently made toward the evening of April 13. Presumably the two stayed throughout the night by their machine, depriving other planes the next day on the long walk to Amman, not aware that the city was 180 miles distant.

## Flight Proceeds

In the morning, the other two pilots, proceeded to Basma, Persia, and on the right, one of the planes was forced down April 15, near Samarra, on the Euphrates River, about 150 miles from Baghdad. The other machine proceeded and took several attempts at Basma. The pilots apparently signed the flight as indicating relief from a long period. However, continuing the flight together, they left Basma the next day in their actual and reached Bunder Abbas, Persia, in the evening. On April 16, the two planes continued on to Basma, where they reached, but due to the distance between Basma and Kermah being about 725 miles. The next day of the flight was commenced at 7:15 A. M.

on April 16, for Agna, India, and the 800 miles was covered in 2 to 10 min., Agna being reached at 2:45 P. M. In the meantime the search for Captain Rodriguez and his mechanics. Agna was contacted and on April 17 the report was received that Captain Rodriguez was found by Flying Officer Coghill, of the Royal Naval Air Force, sixty miles from his abandoned machine. His feet were swollen, but otherwise his condition was good and he was taken back by step to the nearest medical station. Meanwhile Agna was found twenty-five miles further away from the plane. This member of the three added by the incredible risk of being flying across the desert, went to leave the machine in the center of a forced landing. The two were immediately taken to hospital in Agna, although the condition of Agna, who had gone on ahead of his companions, Rodriguez, in the hope of finding relief in the desert, was such that he could not be moved immediately, his condition being so bad.

It is understood that after leaving his companions, and re-entraining Rodriguez, was forced back to the Baghdad station. Captain Rodriguez was forced to land, owing to lack of fuel. His machine was found five miles outside the track actually followed by traversing the desert.

Captains Lopez and Godoy continued their flight and, on April 18, reached Calcutta from Agna, a distance of approximately 700 miles. Lopez was troubled with head fever at Agna and, therefore, left that point about two hours after his companions. Apparently most of the flying on the tropical route was completed at about 6,000 ft., being found to be too high, there. The last was very extensive. After

a week at Calcutta, the three reached Bangalore, Bharu, on the afternoon of April 25, having left Calcutta in the early morning and passing through very bad weather.

More attempts overtook the flight on May 1, when, after Lopez and Godoy left Basra, Persia, India, Calcutta, together, Lopez and his mechanics, Calcutta, were not heard of. Calcutta arrived at Basra on Saturday, afternoon, May 1, and damaged his machine in landing. Plans are reported that possibly Lopez has been saved since the view in the Gulf of Persia on the South China Sea. Godoy's companion lost most of the 500-mile flight, chiefly poor water, was difficult because of misty weather, making visibility extremely poor.

At the time of this going to press, there is some doubt as to the possibility of the flight being continued, since if Lopez is found. The American Government originally refused to permit the Spanish fleet to land on a run by over the Gulf of Persia, on the flight between Basra and Manila. Lopez, however, it is reported that the Japanese government is to send three machines in the event of reaching the plane, as soon as the island on condition that under a reconnaissance of the route, covering a landing field, be made only in case of extreme emergency. It is believed, however, that the flight, under such conditions, would be extremely hazardous, if being started that it would be absolutely necessary for the planes to land on the island of Formosa in emergency landings. The distance from Manila to Moscow, direct, is about 10,000 miles. No Indian ship, as yet, has been found, nor has Captain Lopez been heard of since.

## A Danish Long-Distance Flight

A flight by Danish planes, starting from Copenhagen in Japan is being planned for the Spring. One Danish-built Fokker, with Latécoere Dornier 400 hp. engines will be used, and the route will probably be across Europe and via India and China.

## S.A.E. Aeronautical Meeting in August

Plans are being made for a two-day aeronautical engineering meeting to be held at Philadelphia by the Society of Automotive Engineers, just before the National Air Show there. The date selected previously will be Aug. 30 and 31, and the place, the Delaware State Hotel. Considerable attention will be given to the addresses and discussions to the subjects of design, construction and questions of both technical and higher-class craft.

As the meeting will be held at a time when the Society's annual convention is in full swing, there will be a triple session for the top airfield where this is expected.

## The National Balloon Race

The second round of the National Balloon Race which is held at Little Rock, Ark., is now in progress. While Ward T. Van Cleave, flying the "Goodwin 11," won the first, the second and third places are to be won by the Goodwin 11, flying the "Goodwin 11," and the "Goodwin 11," flying the "Goodwin 11," respectively. Preliminary reports from the first two races are in the reverse order.

Year of Pilot and Airplane	Pilot	Plane	Year of Balloon	Location	Distance
1. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
2. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
3. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
4. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
5. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
6. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
7. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
8. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
9. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100
10. W. T. Van Cleave	Goodwin 11	Goodwin 11	1937	Little Rock, Ark.	100



The "Goodwin 11" (left) and the "Goodwin 11" (right) about to start. Little Rock, Ark., on the start of the National Balloon Race.

The second National Balloon Race, starting at Little Rock, Ark., is now in progress. Preliminary reports from the first two races are in the reverse order.



The first Hispano XLII (Latécoere-Dornier 400 hp. engine) just before the start of the Madrid-Manila flight.



## Army-Navy Standards

**Standardization of Aircraft Parts, Undertaken Jointly by Army and Navy, Will Enable Closer Cooperation from Procurement Standpoint. A Great Advantage to the Aircraft Industry.**

By MAJOR LESLIE MACDILL, A.S. and LIEUT. RALPH S. BARNABY, U.S.N.\*

THE ARMY AIR SERVICE and the Bureau of Aeronautics, War Department, are now making the first of a new series of standard part drawings. These drawings, known as AN standards, represent parts which have been standardized by the Army Air Service and the Bureau of Aeronautics. The standardization of the parts of a good standard, consisting of representations of the two Services, approved by their respective chiefs for this purpose.

The drawings held by this committee are issued regardless of the common airplane manufacturing companies, manufacturers of materials and parts for aircraft, and the Aeronautics Standards Committee of the Society of Automotive Engineers. This way, the two services have represented, and it is implied due to their cooperation and assurance that the AN Standards have become possible.

### War-Time Problems

The advantages of such standardization are manifold. During the war, when the Services were buying airplanes and parts in large quantities, it was necessary that standards be adopted rapidly. Naturally the results obtained were different. These differences have persisted, and, as a result, as Army airplanes, such as the C-1, have been just a little different from a Navy part used for the same purpose. Different engines, however, so that a Navy inspector could not accept the Army part, and vice versa. This required the part manufacturer to produce and carry in stock two kinds of material, and for both Services to purchase and maintain two separate stocks. From the service point of view, it represented increased cost and prevented exchange of material between the two Services. It is the purpose of the AN Standards to eliminate these conditions.

The Society of Automotive Engineers, through its Aeronautics Standards Committee, is adopting these AN Standards as recommended practice for civil aviation.

Naturally, changes on such a large scale must be adopted gradually, and as is effected in a certain measure by the standard and standard list. It is believed that the Navy's method of accomplishing this is explained clearly in the following excerpts from a Bureau of Aeronautics letter accompanying the AN drawings, addressed to Inspection of Naval Aircraft and to the Naval Aircraft Factory:

"1. As a result of successful liaison between representatives of the Army Air Service and the Bureau of Aeronautics, an attempt is being made to standardize the specifications and drawings for standardized material and fittings. It is hoped that, at this time, this will include all materials, the so-called standard parts and fittings, and equipment common to the two Services.

"2. The resulting specifications and drawings will be known as AN or Army Navy Aircraft Standards. In the case of the drawings, it is intended that they represent the corresponding Naval Aircraft Factory drawings in the Standard Catalogue. The AN specifications will serve as master specifications for the preparation of new Navy Department Letter Specifications for the material or parts concerned.

"3. The facilities for the use of the AN Standard Drawings as a reference list has been proposed, showing the corresponding AN number for every Naval Aircraft Factory part number for which an AN standard exists.

"4. It is requested that the AN Standards be inserted in the front section of the Standard Catalogue.

and that the corresponding Naval Aircraft Factory drawings be shown on the last blank immediately preceding the standard drawing. This will enable the manufacturer to check the standard or parts as well as to the standard manufacturer. With this in mind, it was considered to keep the new AN parts interchangeable with the old whenever possible.

"5. One of the necessities, in developing the AN Standards, was that their adoption should not work to cause loss or difficulty to the manufacturer, an amount of stock of old standard parts and materials. This was taken into consideration in the standard or parts as well as to the standard manufacturer. With this in mind, it was considered to keep the new AN parts interchangeable with the old whenever possible. Accordingly, it was agreed that the presentation of the AN Standards would not prevent the airplane manufacturer from using up the old material which they have in stock or may be required to purchase in the near future.

"6. This last provision is necessary in order to work to reduce hardship on the material or parts manufacturer. For this reason, it was decided that he should be permitted to carry on an inventory of old standard material or parts conforming to the old measurements until his supply is exhausted, unless, by special agreement between manufacturer and purchaser, it is agreed to furnish the AN parts. It is intended, however, that the above apply only to parts manufactured before presentation of the AN Standards, and that, in the future, such parts be made in conformity therewith.

"7. Inspection shall not be to that, after the date of presentation, new stocks of parts are manufactured in accordance with the AN Standards for use on Navy Standards.

"8. The date of presentation for the standard drawings included in the present list is July 1, 1936."

### New Air Service Standard Book

The Army Air Service has issued a directory at this time, to serve as a directory new Standard Book which will include the AN Standards. This book is distributed to all Air Service Construction and Inspection. The preface to this book contains the following paragraph, which outlines the attitude of the Air Service regarding the use of old standard parts:

"The issue of this new Standard Book does not automatically exclude any stock on hand listed in the old book but not included in the new one. The old stock will be used for replacement in the Air Service equipment already in use and may be used as new articles, until the stock on hand at contractors' plants is exhausted.

"Old stock carried by members in the New Standard Book is to be identified on new drawings by the new part number. This book made up to the standard new drawings to be submitted on order specifying the new numbers until the supply on hand at the part manufacturers plants at the time of issue of this book has been exhausted.

It may be easily realized that the field of AN standardization is almost without limit. The work started in an attempt to do away with the differences between the material specifications of the two Services. It was now apparent that there was only a small part of the subject. The last most obvious field was in the so-called "standard parts" such as bolts, nuts, pins, washers, etc. These fields, and the field of power plant equipment and fittings, have only recently been taken over and are being taken up by the Air Service. With the presentation of the first AN Standard Drawing, however, the ground has been broken and the work should advance rapidly from now on.

## The Pitcairn Fleetwing

**A Five-Seater Passenger Plane With a Curtiss C-6 Engine.**

THOUGH AT present only upon a comparatively small scale, but regarded as a basic standard and as though as are the largest and widest aircraft manufacturers in the country, Pitcairn Aviation, Inc., at Dayton, Ohio, Philadelphia, Pa., must be considered to have taken its place (no doubt) in the aircraft industry. A visit to the company's field in Philadelphia reveals the extensive activity which is being carried on. The biggest, a large one, is in the construction of various types of aircraft. A visit here reveals a vast plant, with a large number of various types of aircraft, while all of them are in production. The company provides space for the storage of its own and other aircraft parts.

In the airplane shops, which are in the process of construction, is being carried on. The work is necessarily well done. Not only is the particular design of the airplanes extremely interesting, being characterized by the manner in which single strength of the structure is obtained, while maintaining general simplicity throughout, but the actual working operation is carried out with a skilled hand and the parts appear to be of the very first order.

### An Interesting First Development

In the modernizing shop, wings and tail sections in a standard model of the company's first design, the Pitcairn Fleetwing, are being assembled. The machine, originally intended as a passenger bus "ship," has proved an acquisition. The design of the machine is of welded steel tube construction, while the wings and tail surfaces are constructed of wood. A close inspection of the wing construction reveals the same close workmanship as is characteristic of the tail fuselage work. The structure is generally carried out along conventional lines. The method of building the tail spine is, however, of special interest. The spine is built up, to suit it generally, of two more or less square spars side by side and forward into one suspended spar with plywood walls. The result is an exceedingly strong structure, possessing, at the same time, marked simplicity and lightness.

Not only are the parts of the main wings of the Fleetwing constructed along this principle, but the ailerons, tapered out of the tail plane is built along the same line. The rigidity

of the complete structure at wing and tail plane is very marked. This was proved when an employee was made to swing the machine structure of a Fleetwing airplane.

Apart from the routine manufacturing activities in the production of further Fleetwings, there is evidence of much creative work being undertaken, not only in the way of original airplane design itself, but in the development of sophisticated methods of various classes, and manufacturing methods. The result has been looking drawings often possess the atmosphere of very distinct industry under the able guidance of Arthur H. Latham, chief engineer to Pitcairn Aviation.

The flying field of the company is a very fine one and another has been begun on the Dayton-Pike, and is said to be the largest commercial field in the East. This field is to be used in the passenger-carrying situation and two large hangars are already under construction, one of which will be let out to local operators. This field, without a doubt, will be large enough to handle the business, when the Fleetwing is in full swing, and it will be considered that Pitcairn Aviation has been given the role passenger-carrying rights in the East. The work is being to keep them in the passenger-carrying market and chief pilot of the company, together with the company's other planes, are being shown the business.

Harold F. Pitcairn, himself, shows a very modern standard of design and building. The Fleetwing is such a success in its ability to hold its weight of weight, side of each side of each. A flight in the plane is an interesting experience. The two passenger double seats are very roomy and, at the same time, they are delivered from the seat by the large wood structure. The plane itself is remarkably steady in its own, even on a material heavy day. A pleasant description of the plane will be given.

### The Pitcairn Fleetwing

As originally conceived, the Pitcairn Fleetwing was to be used for the purpose of carrying passengers on short flights. While this purpose has been kept foremost in the minds of the designers, it was also deemed advisable to fit the plane to be used as a short-range commercial machine for emergency passenger, express or



The Pitcairn Fleetwing (Curtiss C-6, 160 hp. engine) five-seater passenger plane

\*Major MacDill is Chief Engineer, Inspection Division, MacDill Field, Tampa, Fla.; Lieut. Barnaby is Chief of the Construction Group, Bureau of Aeronautics, Washington, D. C.



at Bryn Aulis early in 1935. Kiehl for its rapid climb and slow landing speed, the plane proved its performance during flight tests conducted by Carl Feltner, Executive Secretary, United Commission of the National Aeronautical Association, and C. Townsend Macdonald, Pennsylvania Governor of the National Aeronautical Association, who officially certified its performance figures after tests on April 17 and 18, 1936.

The details follow:

(Certified by National Aeronautical Assn.)

Light with pilot, oil, water, and 32 gal. gasoline and with passengers

Maximum speed at ground 158.8 m.p.h.

Cruising speed at 10,000 ft. 135 m.p.h.

Landing, with pilot, oil, water, 32 gal. gasoline and with 700 lb. of mail

(Total weight 2861 lb.)

Maximum speed at ground 164.2 m.p.h.

Cruising speed at 10,000 ft. 135 m.p.h.

(Total weight 2861 lb.)

General Characteristics

The general features of the machine are as follows:

Span, upper wing 38 ft.

Span, lower wing 32 ft. 3 in.

Wing, both wings 62 sq.

Gap between wings 6 ft. at center

Blades 25 in.

Inverters, upper wings 35.4 gal.

Inverters, lower wings 35.4 gal.

Dihedral, upper wings 6 deg.

Dihedral, lower wings 5 deg.

Length overall 52 ft. 13 in.

Wing, upper 9 ft. 16 in.

Wing, lower 10 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

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Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

Wing, total 20 ft. 10 in.

as made. Prior to its connection with this company, Mr. LaPrie had spent considerable time for the Navy flying out of Chicago, but several years. Previous to that time he served and operated several Curtiss biplane flying boats at the Edgarwood Beach Hotel. He also had night flying at the Chicago Program of Progress held at the Municipal Pier 1918 and 1923.

Mr. Curren, the sales manager, has a wide reputation as an engineer and is capable of handling any, having been connected with some of the largest engineering work in the United States.

### Artificial Fog Dispersion Experiments

One of the greatest reasons to date aerial navigation—over the time of every minute may be greatly lessened by the successful completion of experiments being conducted by the Naval Bureau of Aeronautics in the propagation of fog over airplane landing fields.

Although the experiments, which have been in progress for several years, have not been completed, they have reached the point where confidence is expressed that fog-propagation apparatus can be used in dispersing of fog-banks over landing fields, thereby clearing paths in which planes may land in safety.

By means of a specially mounted airplane propeller, electrically charged air has been projected in such fashion as to simulate an electrically charged curtain of air which penetrates the fog and then opens a clear path for landings. Time experiments have been carried on at the Naval Air Station at Dayton, Ohio, and at the Naval Air Station at Pensacola, Fla.

Tests conducted with the experiments indicate that 375,000,000 ft. per minute of fog drifts at the rate of 2 m.p.h. over a landing field through a vertical mass of charged air having a radius of 1,000 ft. with the center on the ground.

One type of apparatus with which the Navy is experimenting is capable of passing through a 700,000 ft. of air a minute, which it charges electrically and throws into charged curtains. It is estimated that these curtains can cause the precipitation of about 95 per cent of the fog moving over the landing field, thus clearing a radius 1,000 ft. high and 2,000 ft. wide over the full length of the landing field.

The apparatus consists of a simple type of engine charging system, a transformer with rotating apparatus and an airplane propeller and engine all mounted on a truck. It is now demonstrated that an airplane propeller driven at 4,000 ft. per minute mounted on a truck has sufficient power to project electrically charged air over a vertical plane having a radius of 1,000 ft., thus forming an electrically charged curtain through which the fog cannot enter the landing field and pass.

Important progress in fog penetration has been demonstrated by the Navy in this experiment. It has been found that the wind is electrically charged and that a projected electric field, to drive the fog particles together, is essential for electrical penetration over an open field.

Further experiments will be carried on to determine such points as the distance to which the electric field can be projected, whether the electric current should be set up by direct or alternating current and the proper voltage to use. Other fog penetration experiments are being carried on jointly by the Navy Bureau of Aeronautics and the Air Service on the principle of electrically charging dust or sand particles which are released into the fog from airplanes, thus causing condensation and ultimate dispersal of the fog. The successful completion of these important experiments will remove one of the greatest reasons to date aerial navigation. The field for the use of fog penetration apparatus is wide and so much aircraft will be used to great relief to the traffic congestion and danger on foggy days, and also to harbor and channels where shipping bottoms are congested and delayed by reason of fog.

The two experiments have been carried on by the Navy partly upon an experimental laboratory basis with the object of separating the factors of the subject, the work is now being continued with a view to developing the principles involved in such a manner as to make it possible to utilize the dispersed apparatus upon a practical basis.

## Model Designation of Naval Airplanes

A Full Statement of the System Employed in Designating Naval Airplane Designs.

The particular system of letter designation of airplane designs and numbers for the Navy differs from that employed in it is thought that on flight are the agencies governing the making of the system and to both engineering and military—engineers.

Naval models of airplanes are divided into two classes in accordance with the function to be performed. Each class is designated as follows:

Class	Designation
Observation	O
Transportation or general utility	T
Reconnaissance	R
Training	Tr
Target	Ta
Patrol	P
Reconnaissance	R
Reconnaissance	R

### Class Designation

The letter "N" signifies a heavier-than-air craft or "B" signifies a lighter-than-air craft. The letter is omitted in the designation of observation, reconnaissance, training, target, patrol, and transport models, but must be employed in all official correspondence relative to classes of airplanes, naming of airplane squadrons, etc. When an airplane performs functions of more than one class, the primary function is indicated in the designation. Present provision is to designate airplanes by a group of letters and numbers. This consists first of a letter indicating the class, second, a number indicating the model and third, a letter indicating the manufacturer, followed by a dash and a number indicating the modification of the model.

Model	Manufacturer	Features	Engine	Remarks
O-1 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-1 (N)	First in class (N)
O-2 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-2 (N)	First in class (N)
O-3 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-3 (N)	First in class (N)
O-4 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-4 (N)	First in class (N)
O-5 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-5 (N)	First in class (N)
O-6 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-6 (N)	First in class (N)
O-7 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-7 (N)	First in class (N)
O-8 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-8 (N)	First in class (N)
O-9 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-9 (N)	First in class (N)
O-10 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-10 (N)	First in class (N)
O-11 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-11 (N)	First in class (N)
O-12 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-12 (N)	First in class (N)
O-13 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-13 (N)	First in class (N)
O-14 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-14 (N)	First in class (N)
O-15 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-15 (N)	First in class (N)
O-16 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-16 (N)	First in class (N)
O-17 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-17 (N)	First in class (N)
O-18 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-18 (N)	First in class (N)
O-19 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-19 (N)	First in class (N)
O-20 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-20 (N)	First in class (N)
O-21 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-21 (N)	First in class (N)
O-22 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-22 (N)	First in class (N)
O-23 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-23 (N)	First in class (N)
O-24 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-24 (N)	First in class (N)
O-25 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-25 (N)	First in class (N)
O-26 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-26 (N)	First in class (N)
O-27 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-27 (N)	First in class (N)
O-28 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-28 (N)	First in class (N)
O-29 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-29 (N)	First in class (N)
O-30 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-30 (N)	First in class (N)
O-31 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-31 (N)	First in class (N)
O-32 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-32 (N)	First in class (N)
O-33 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-33 (N)	First in class (N)
O-34 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-34 (N)	First in class (N)
O-35 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-35 (N)	First in class (N)
O-36 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-36 (N)	First in class (N)
O-37 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-37 (N)	First in class (N)
O-38 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-38 (N)	First in class (N)
O-39 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-39 (N)	First in class (N)
O-40 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-40 (N)	First in class (N)
O-41 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-41 (N)	First in class (N)
O-42 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-42 (N)	First in class (N)
O-43 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-43 (N)	First in class (N)
O-44 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-44 (N)	First in class (N)
O-45 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-45 (N)	First in class (N)
O-46 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-46 (N)	First in class (N)
O-47 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-47 (N)	First in class (N)
O-48 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-48 (N)	First in class (N)
O-49 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-49 (N)	First in class (N)
O-50 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-50 (N)	First in class (N)
O-51 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-51 (N)	First in class (N)
O-52 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-52 (N)	First in class (N)
O-53 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-53 (N)	First in class (N)
O-54 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-54 (N)	First in class (N)
O-55 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-55 (N)	First in class (N)
O-56 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-56 (N)	First in class (N)
O-57 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-57 (N)	First in class (N)
O-58 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-58 (N)	First in class (N)
O-59 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-59 (N)	First in class (N)
O-60 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-60 (N)	First in class (N)
O-61 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-61 (N)	First in class (N)
O-62 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-62 (N)	First in class (N)
O-63 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-63 (N)	First in class (N)
O-64 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-64 (N)	First in class (N)
O-65 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-65 (N)	First in class (N)
O-66 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-66 (N)	First in class (N)
O-67 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-67 (N)	First in class (N)
O-68 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-68 (N)	First in class (N)
O-69 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-69 (N)	First in class (N)
O-70 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-70 (N)	First in class (N)
O-71 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-71 (N)	First in class (N)
O-72 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-72 (N)	First in class (N)
O-73 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-73 (N)	First in class (N)
O-74 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-74 (N)	First in class (N)
O-75 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-75 (N)	First in class (N)
O-76 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-76 (N)	First in class (N)
O-77 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-77 (N)	First in class (N)
O-78 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-78 (N)	First in class (N)
O-79 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-79 (N)	First in class (N)
O-80 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-80 (N)	First in class (N)
O-81 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-81 (N)	First in class (N)
O-82 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-82 (N)	First in class (N)
O-83 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-83 (N)	First in class (N)
O-84 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-84 (N)	First in class (N)
O-85 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-85 (N)	First in class (N)
O-86 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-86 (N)	First in class (N)
O-87 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-87 (N)	First in class (N)
O-88 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-88 (N)	First in class (N)
O-89 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-89 (N)	First in class (N)
O-90 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-90 (N)	First in class (N)
O-91 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-91 (N)	First in class (N)
O-92 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-92 (N)	First in class (N)
O-93 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-93 (N)	First in class (N)
O-94 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-94 (N)	First in class (N)
O-95 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-95 (N)	First in class (N)
O-96 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-96 (N)	First in class (N)
O-97 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-97 (N)	First in class (N)
O-98 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-98 (N)	First in class (N)
O-99 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-99 (N)	First in class (N)
O-100 (N)	Curtis & S. M. Co., 1924	Open TB engine 500 ft. 500 ft.	O-100 (N)	First in class (N)











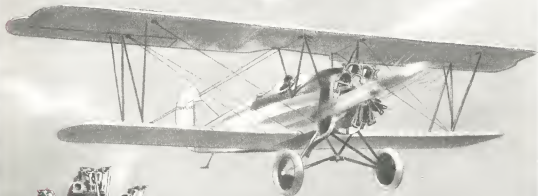






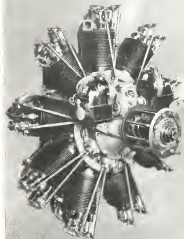
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